Laser Frequency Stabilization for GRACE-II

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- Co-Is:
  - W. Klipstein, N. Yu (JPL), M. Stephens, J. Leitch (BATC)
- Collaborators:
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- Cavity/enclosure development
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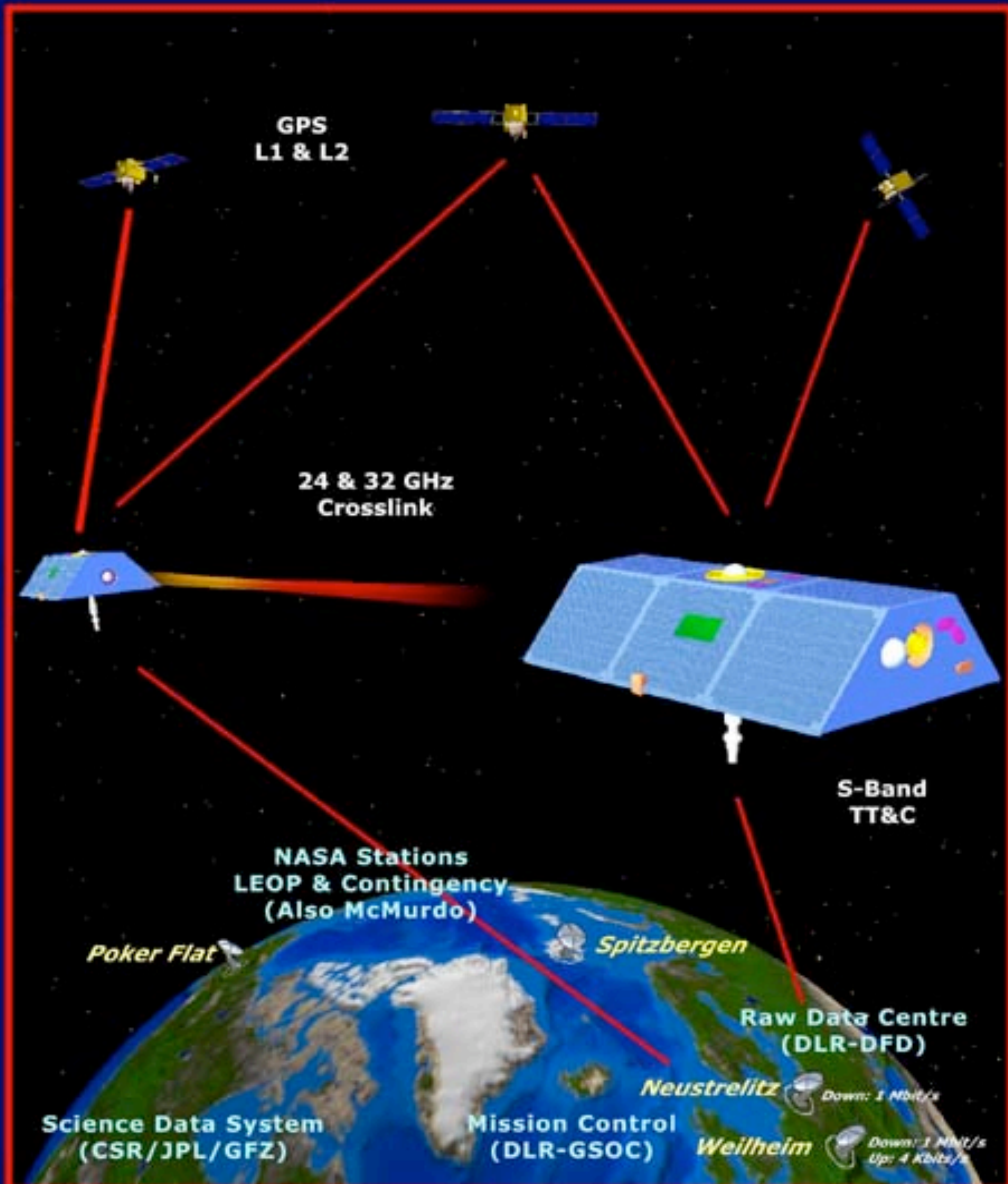


GRACE Mission

Science Goals

High resolution, mean & time variable gravity field mapping for Earth System Science applications.

Mission Systems

Instruments

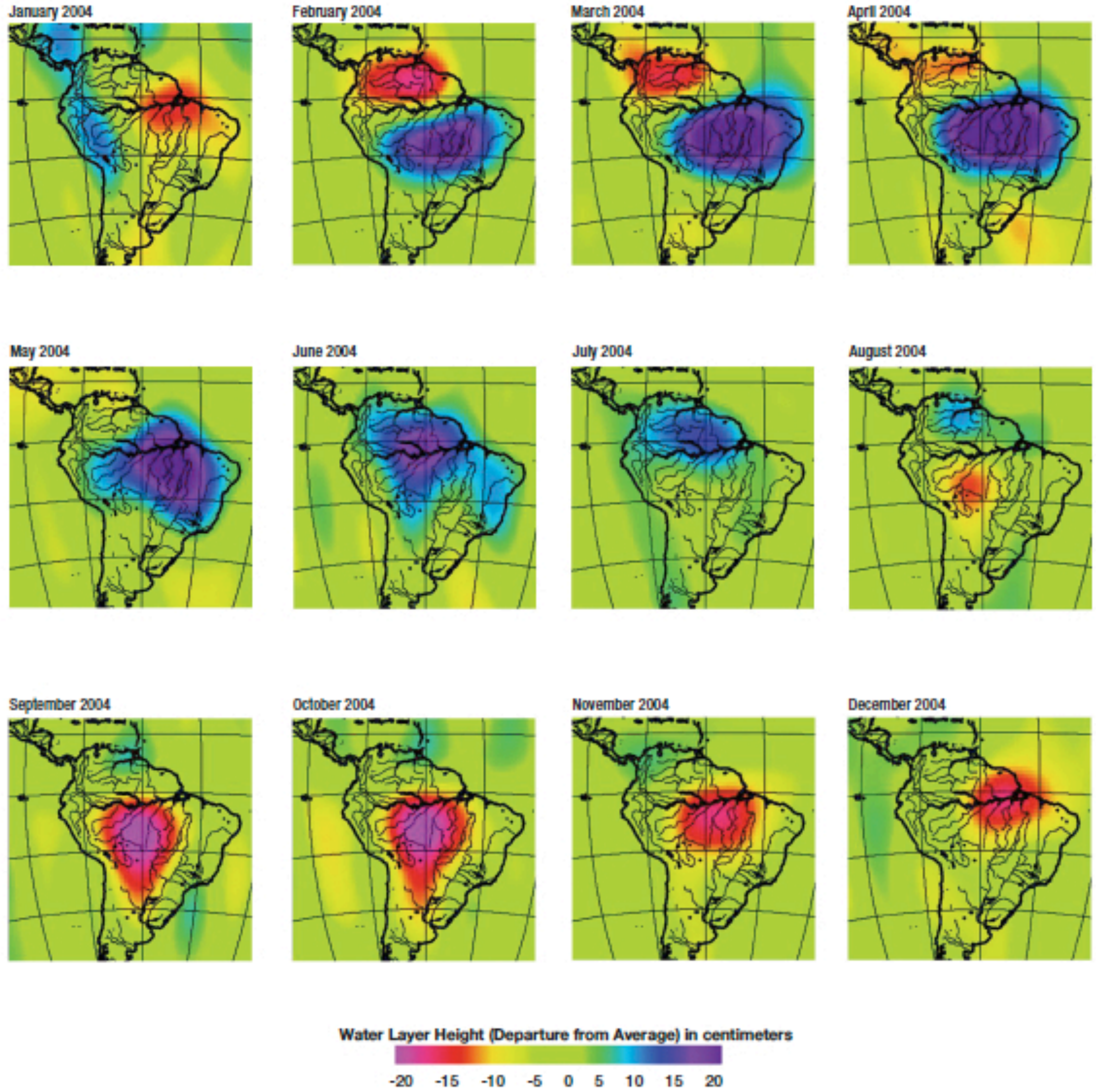
- KBR (JPL/SSL)
- ACC (ONERA)
- SCA (DTU)
- GPS (JPL)

Satellite (JPL/DSS)
Launcher (DLR/Eurockot)
Operations (DLR/GSOC)
Science (CSR/JPL/GFZ)

Orbit

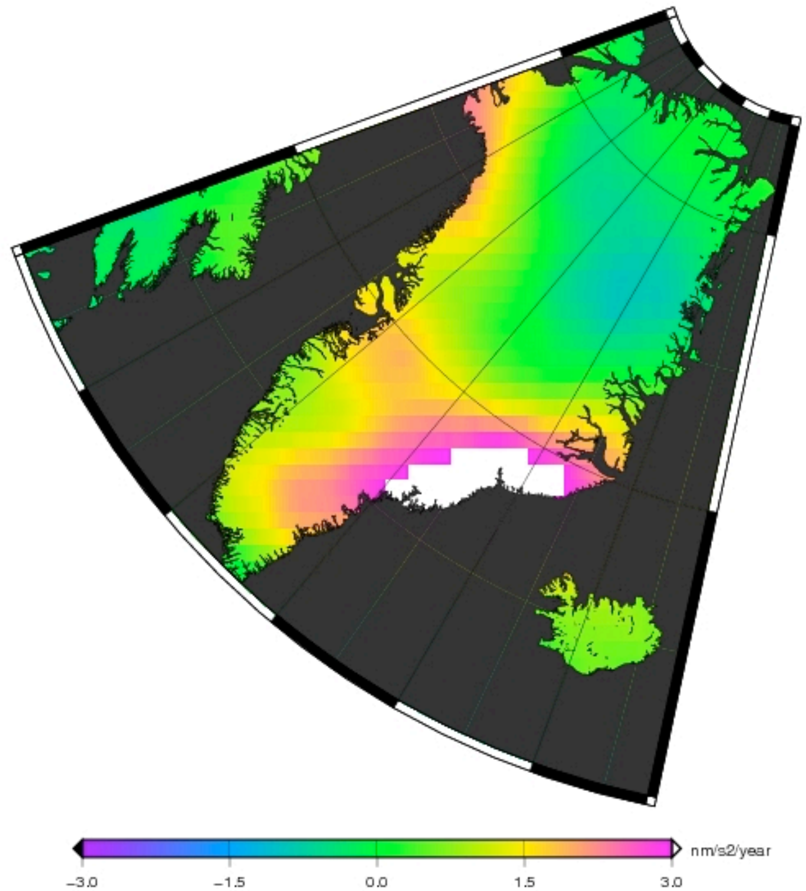
Launch: March 2002
Altitude: 485 km
Inclination : 89 deg
Eccentricity: ~0.001
Lifetime: 5 years
Non-Repeat Ground Track
Earth Pointed, 3-Axis Stable

Seasonal Mass Change in Amazon



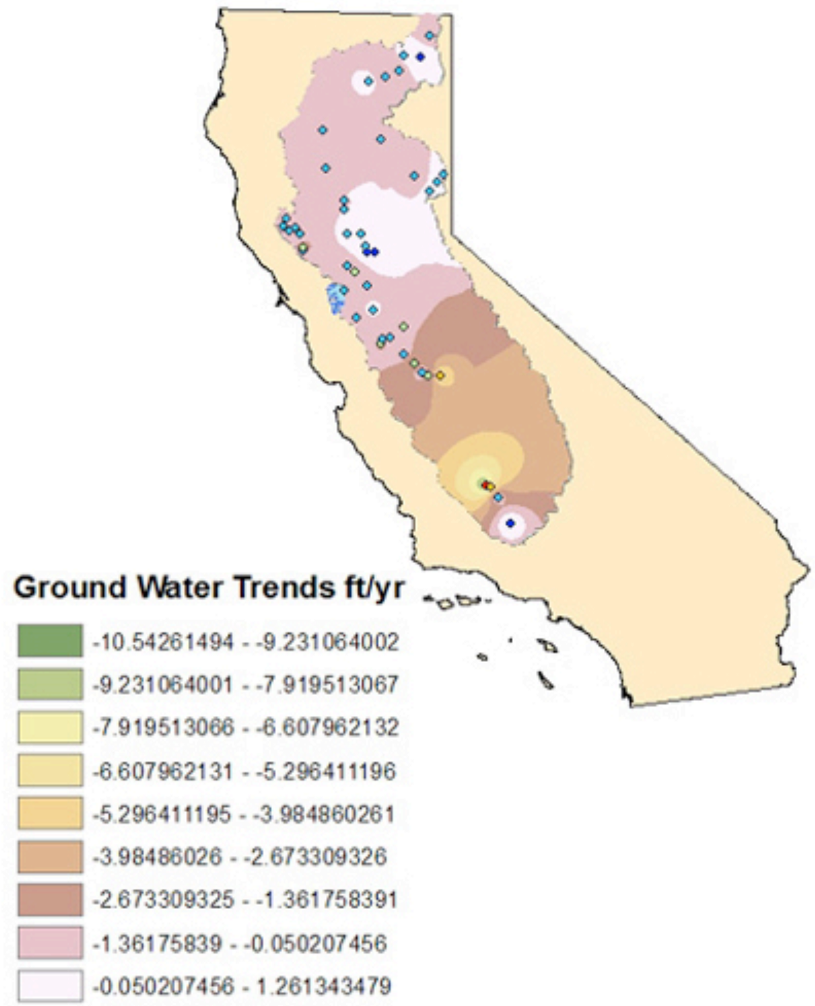
Ice Mass Loss

- GRACE can see changes in mass over large regions
- Effect of most significant features (e.g. glaciers) is spread due to limited spatial resolution
- Improved measurement system will allow better identification of features where mass is changing



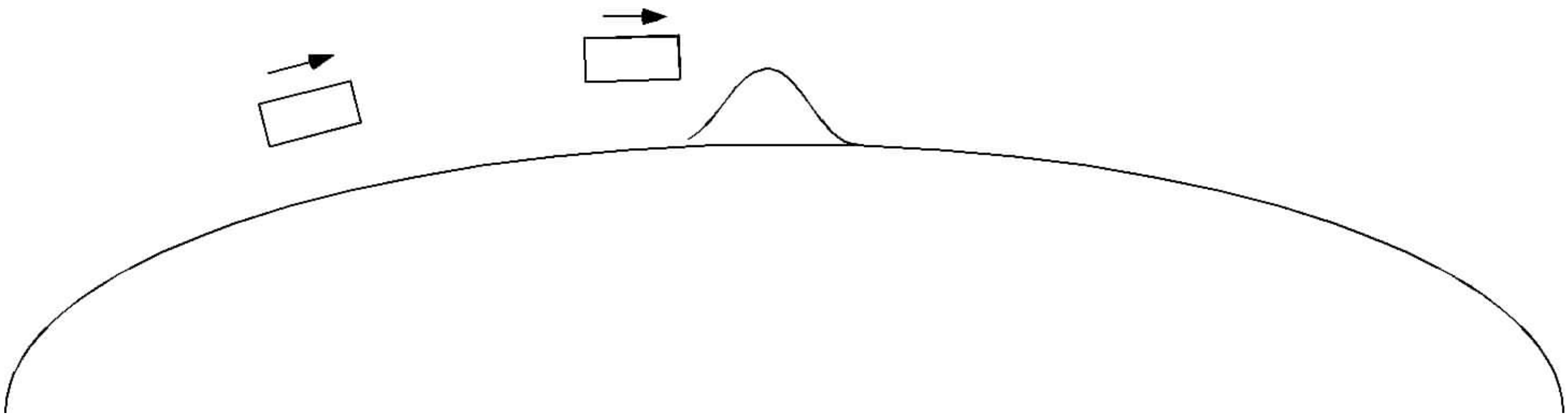
Subsurface Aquifer Changes

- Groundwater storage changes in the Sacramento-San Joaquin River Basins from GRACE and supplementary data,



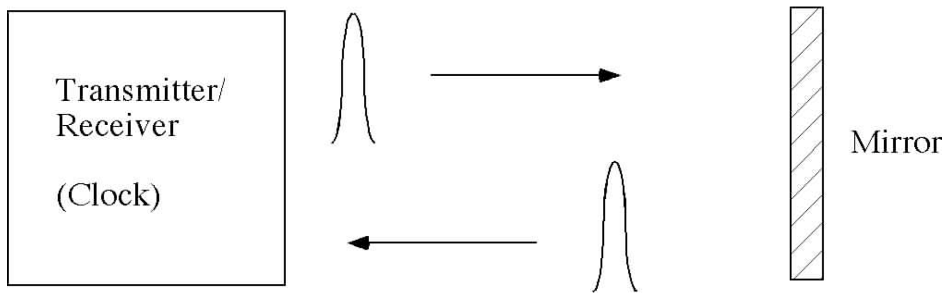
GRACE Measurement Concept

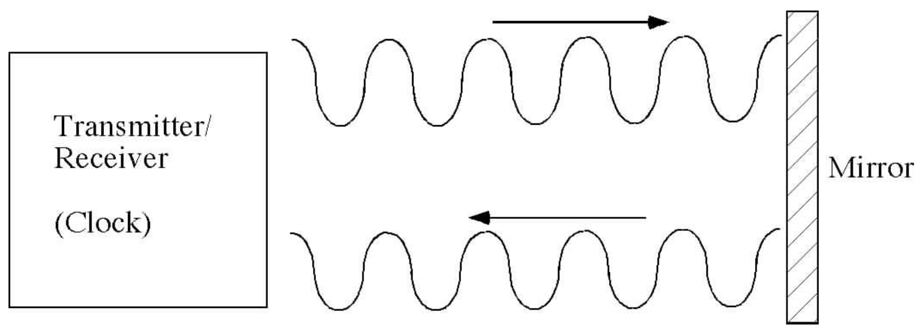
- Earth gravity features affect lead/trailing spacecraft at different times
  - Lead spacecraft encounters feature first
    - e.g. lead spacecraft speeds up towards mountain
      - Range to trailing spacecraft increases
  - Any unknown non-gravity forces acting on spacecraft also affect range
    - Calibrated out using accelerometer or drag-free sensor system



Ranging Measurement Methods

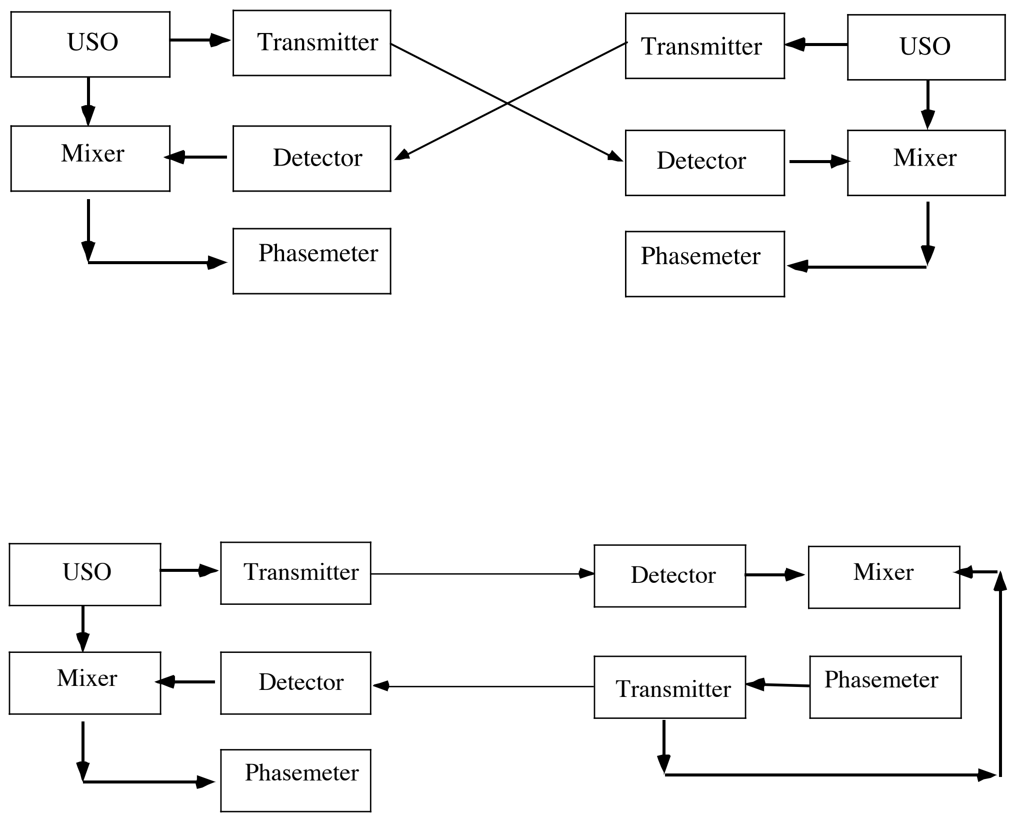
- Range is determined by round-trip light time
  - Pulsed light is used in SLR/LLR where photon rates are low
  - Coherent signals allow use of phase delay for higher accuracy
    - Range ambiguity by integer number of wavelengths can be resolved with modulation if required



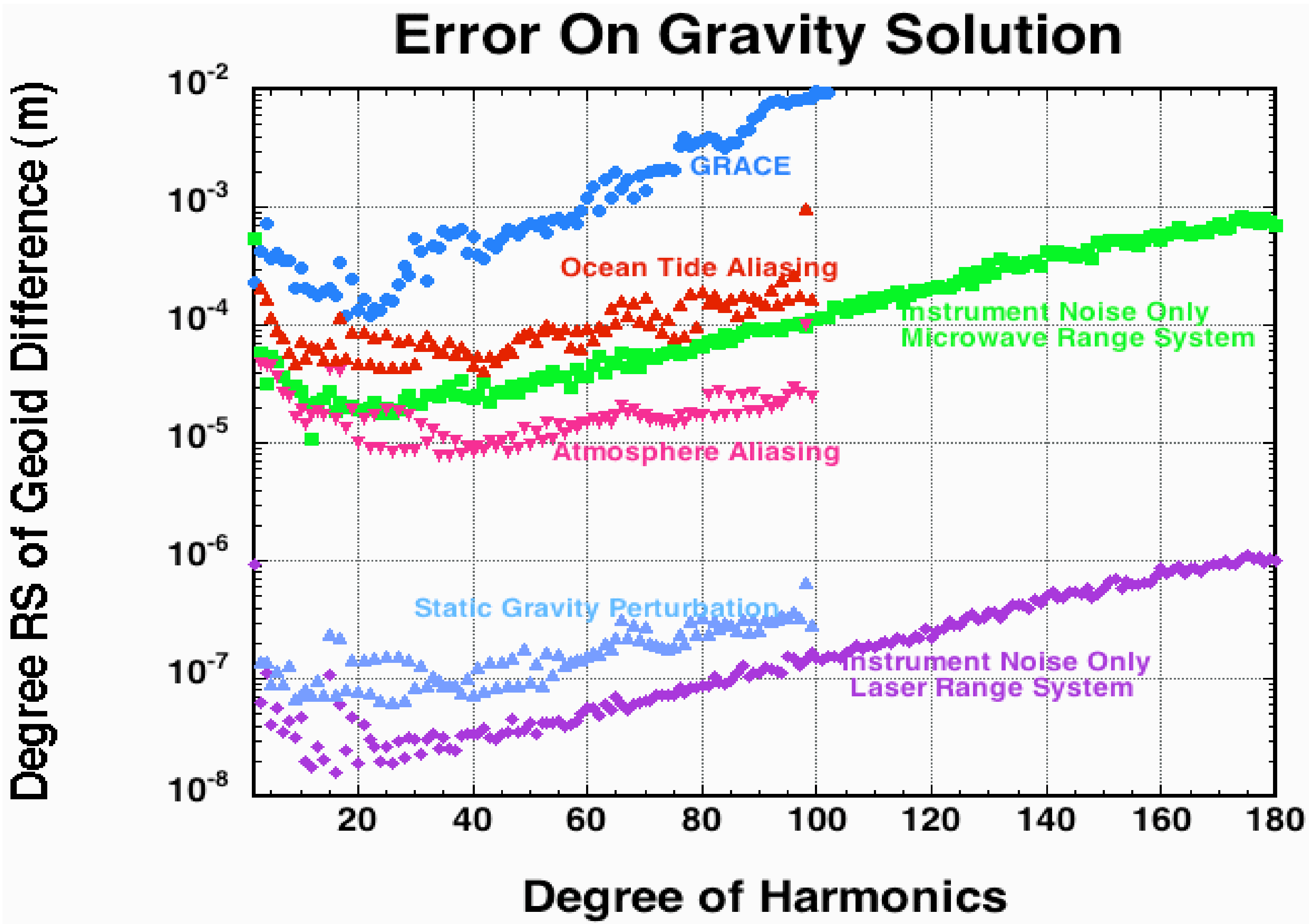


Dual-one-way versus Transponding

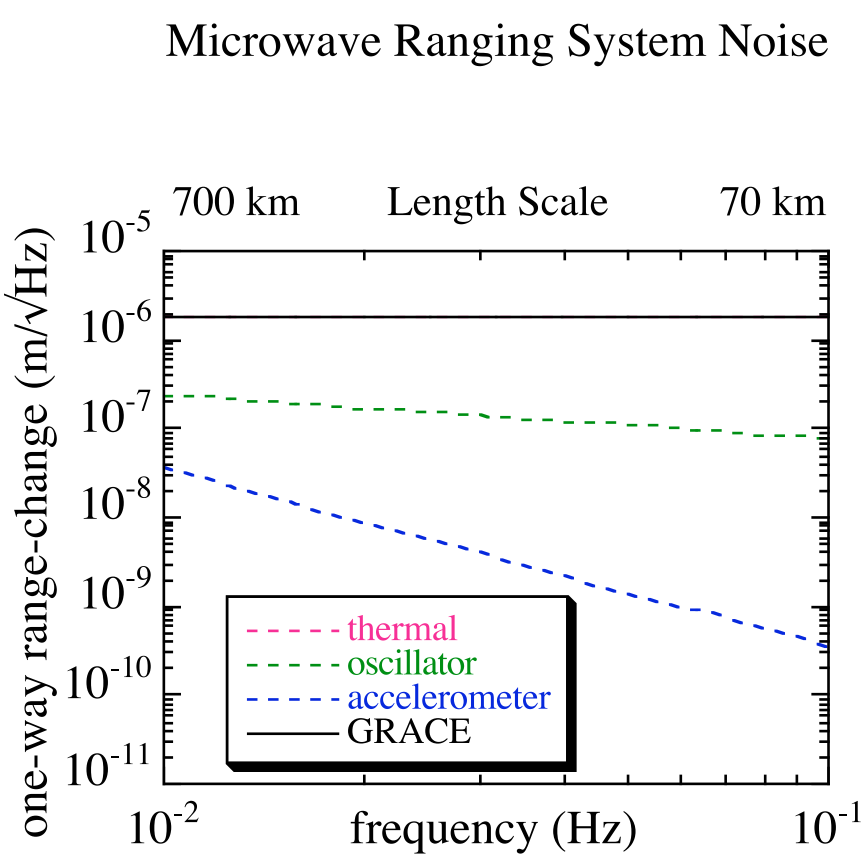
- GRACE uses independent transmission/detection at each spacecraft
  - Combination of data on ground determines range
- Laser ranging will lock laser to frequency reference on one spacecraft and lock laser to received laser on seconds s/c
  - Otherwise laser frequencies would be too far apart
  - Fast phase meter needed for locking function

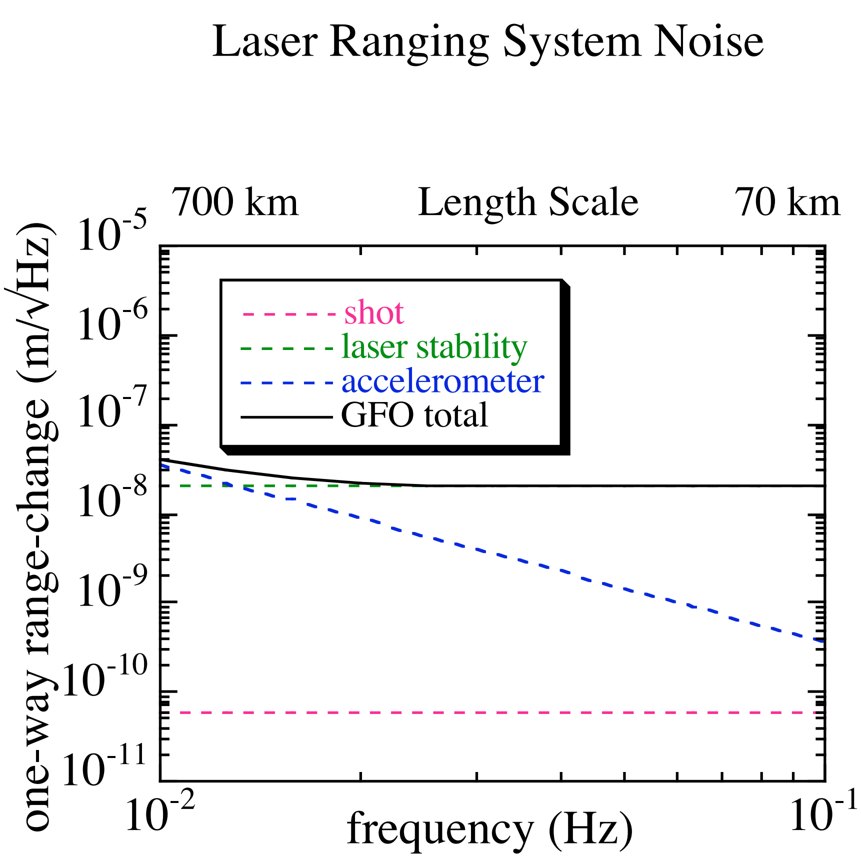


GRACE-II Simulation



Microwave and Laser Ranging Sensitivity



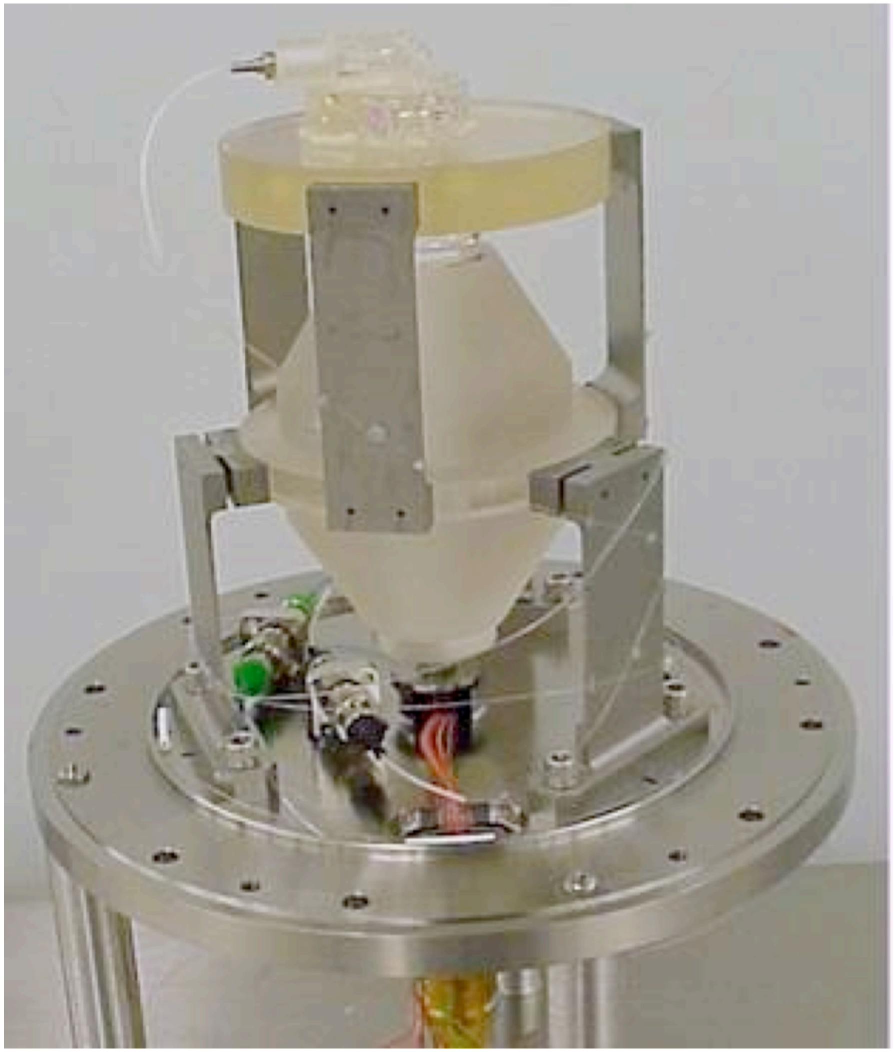


Optical Cavity Frequency Reference

- Stabilize laser frequency by locking wavelength to thermally stable optical cavity
  - ULE glass has ultra-low thermal-expansion coefficient
  - Isolate from external temperature fluctuations

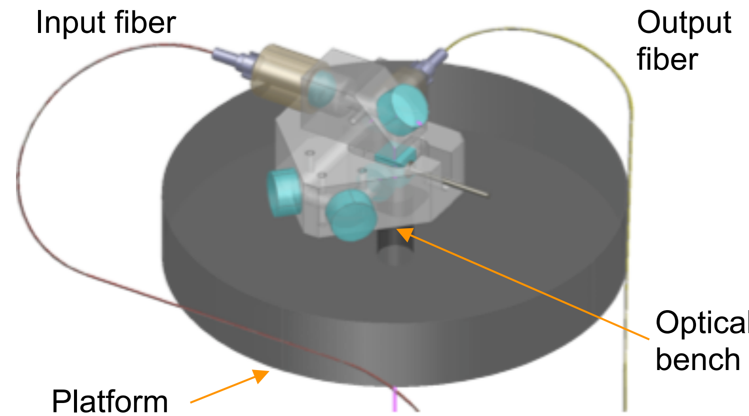
Cavity Mount Design

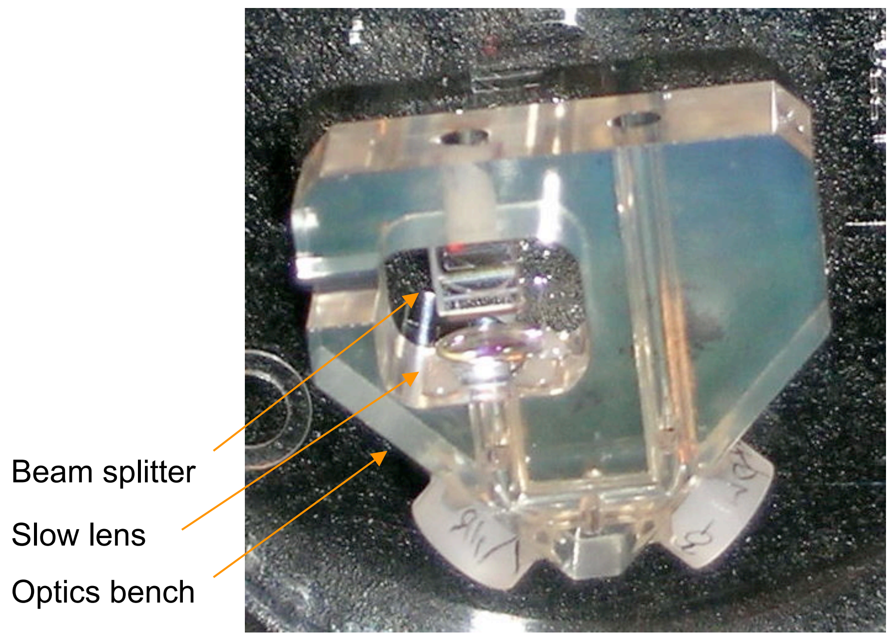
- Cavity is mounted using flexures bonded to cavity central ring
  - Flexure material (Ti) and stiffness chosen to provide support and maintain alignment for launch and minimize thermo-elastic effects on cavity
- Optics for injecting laser light into cavity and output interference via fiber also mounted to cavity using flexures



Optical Fiber Injection

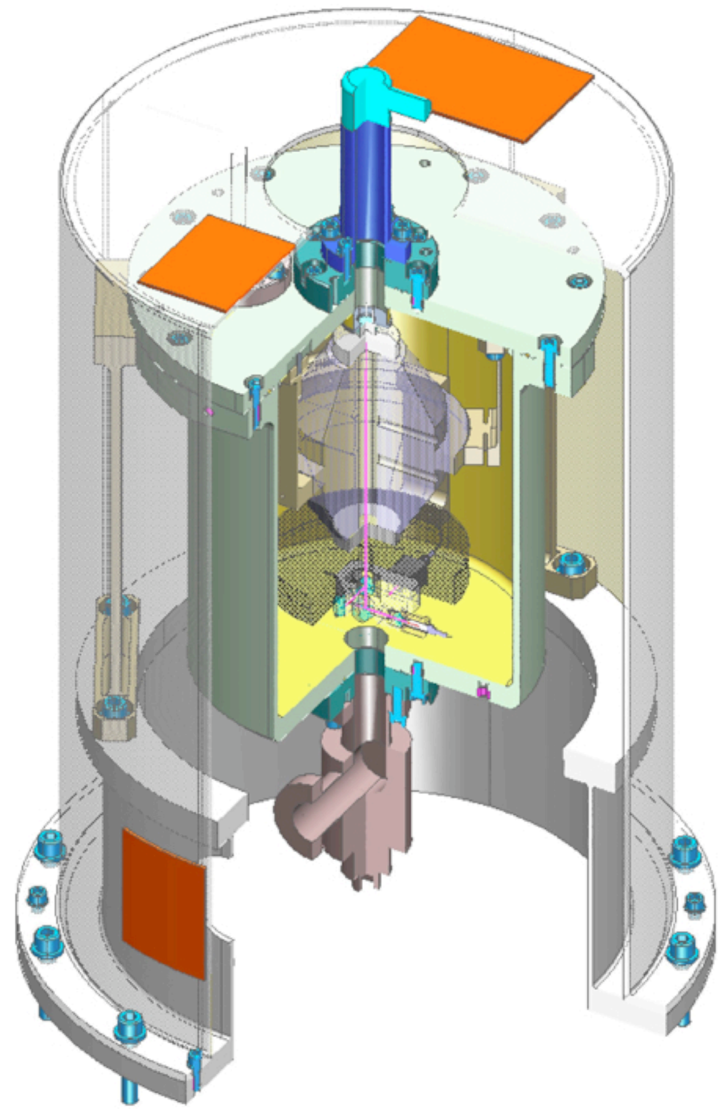
- Laser light injected into, and output signal sent, via fiber
  - –Optics for light injection form small, folded telescope
- Optics, optical bench, and platform are made of low thermal expansion (zerodur) glass and mounted near cavity using flexures





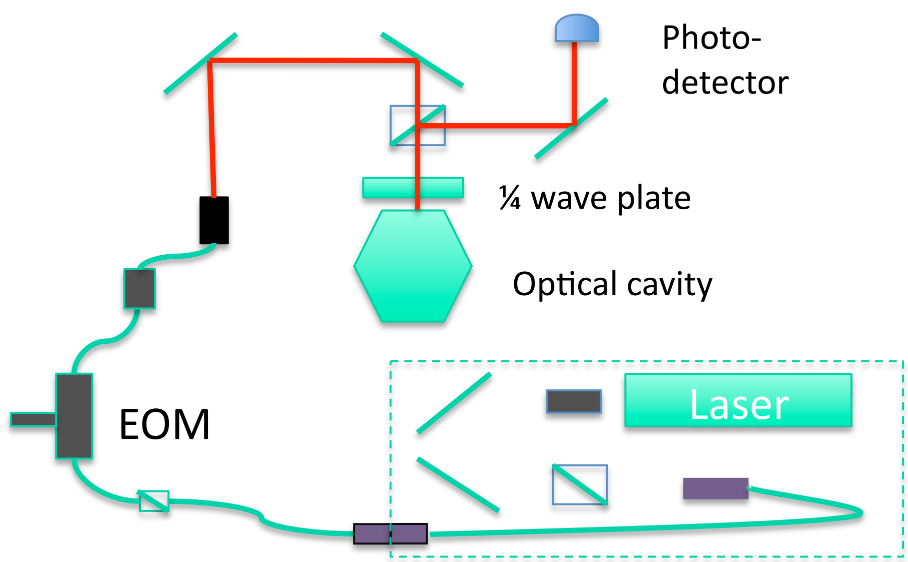
Thermal Isolation Enclosure

- Two concentric shells surround the cavity and optics to isolate them from external thermal
- Size and mass of laser frequency stabilization subsystem has been iterated with GRACE spacecraft team to ensure compatibility
- Inner shell forms a vacuum enclosure to avoid gas pressure fluctuations and contamination
  - Vacuum gauge will allow monitoring
  - Vacuum valve included on breadboard for development, will be replaced with pinch-off for prototype



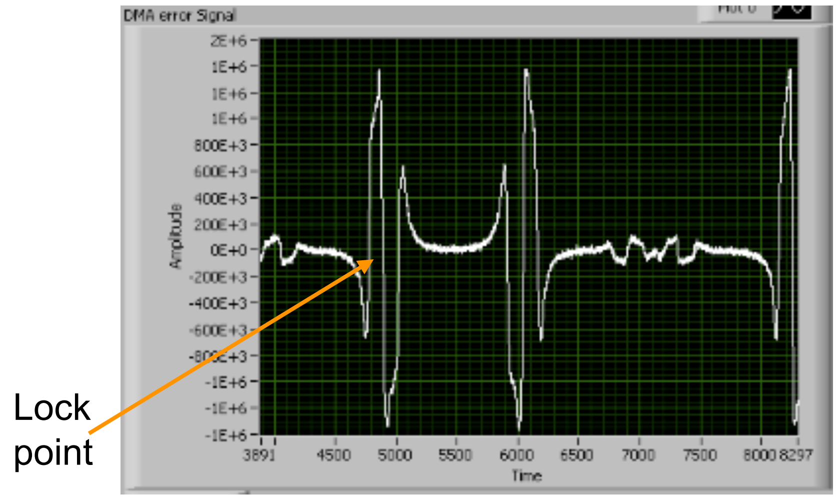
Laser Locking to Optical Cavity

- Laser locked to cavity using Pound-Drever-Hall technique
  - Light resonance occurs when cavity length is integer number of wavelengths
    - Laser frequency stability is directly related to stability of length of cavity
  - Cavity is made of material with low thermal expansion coefficient
  - Light exiting cavity compared with light entering cavity
  - Requires electro-optical modulator to add phase-modulation to laser beam
    - Difference in modulation signal gives laser correction to match cavity

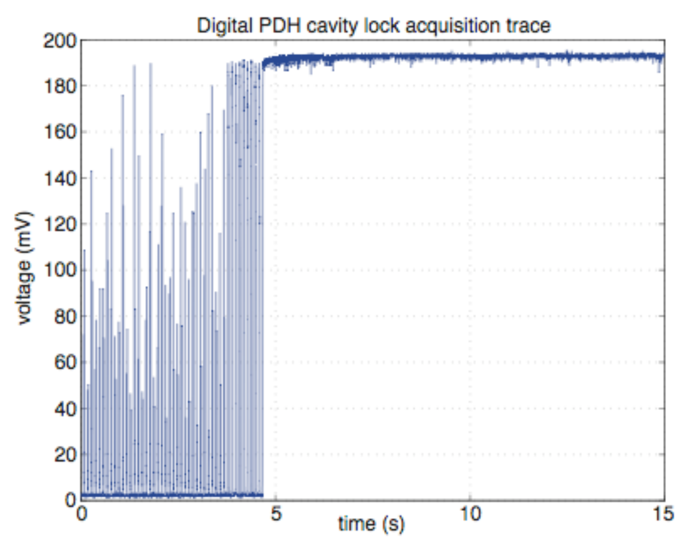


Cavity Response Function

- Laser light injected into cavity exits and interferes with input light
  - –Pound-Drever-Hall locking
- As laser frequency varies at input to cavity, interference with exiting light produces sharply varying response
- When error signal from locking point fed back to laser, laser frequency is stabilized



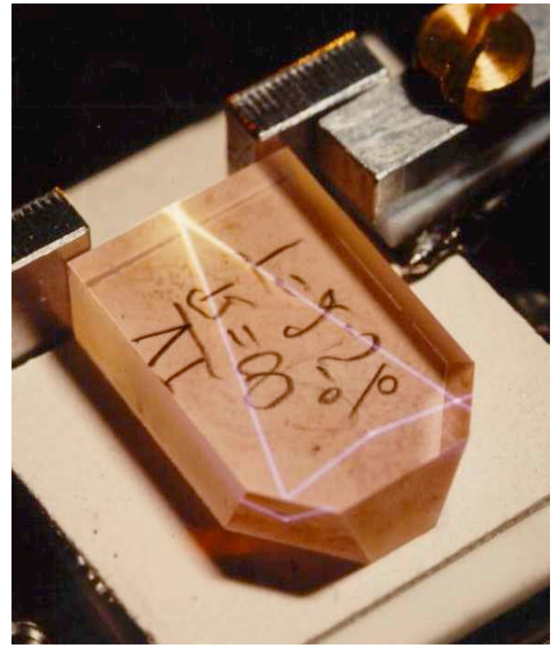
Interference response as laser frequency changed



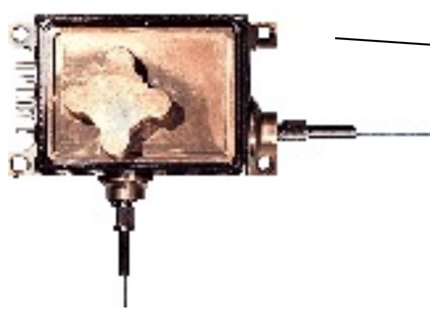
Laser frequency swept than locked

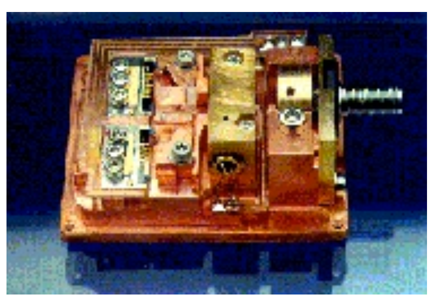
Tunable Laser

- Non-planar ring oscillator (NPRO) Nd:YAG laser provides tunability for locking to cavity
  - Laser wavelength adjusted by changing dimensions of YAG crystal using PZT glued to crystal and thermal adjustment
- Space-qualified NPRO laser available from Tesat Spacecom



NPRO laser head

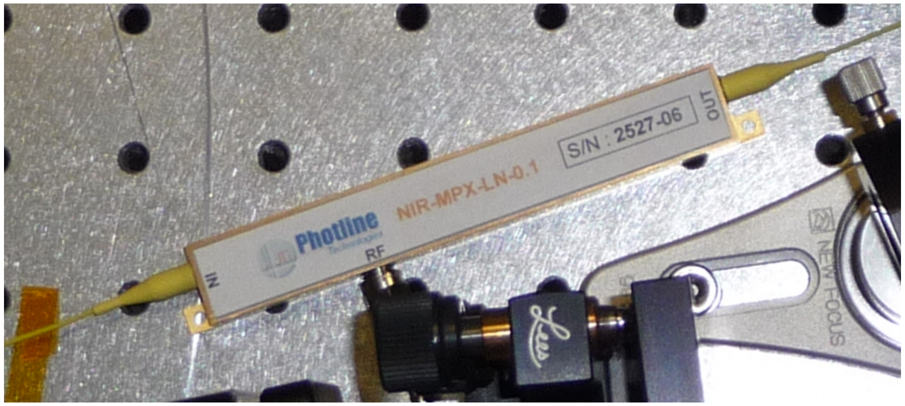




Laser pump diode assembly

Electro-Optical Modulator

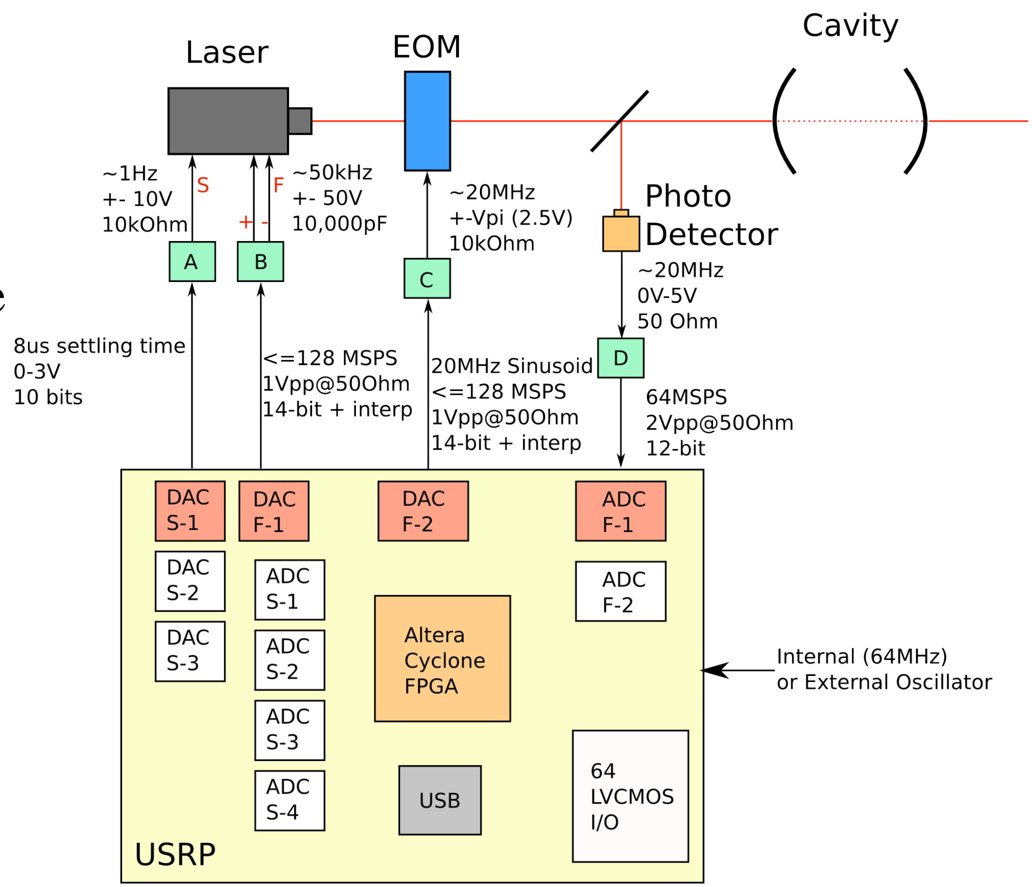
- Several companies make EOM suitable for use in space
- Selected unit from Photline has inexpensive laboratory model traceable to flight model
  - Tested to show meets locking requirements

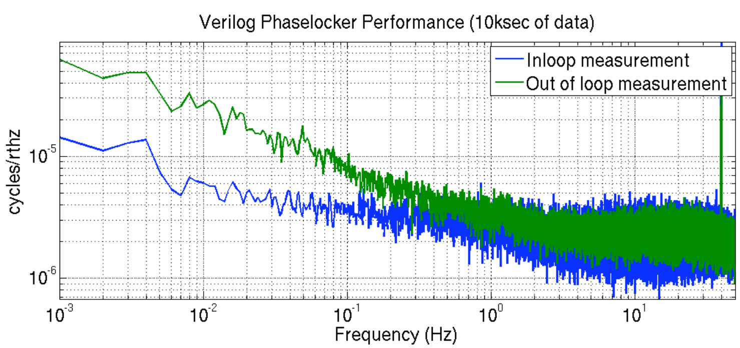


Selected electro-optical modulator

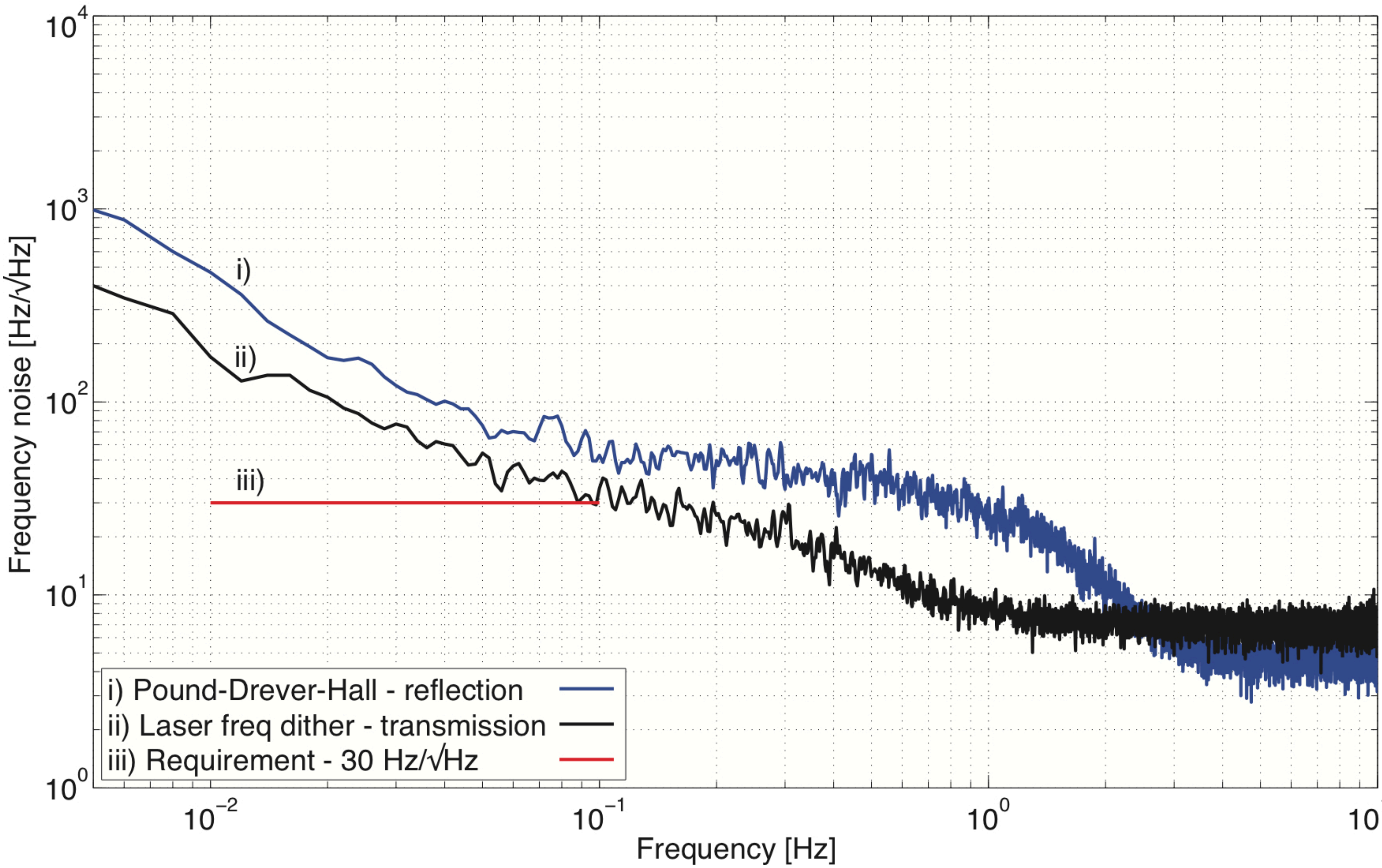
Breadboard Locking Electronics

- Commercial FPGA evaluation board used for development and testing of laser locking algorithms
  - ADC for sampling interference signal from photodiode
  - DACs used to adjust laser crystal frequency via PZT and temperature





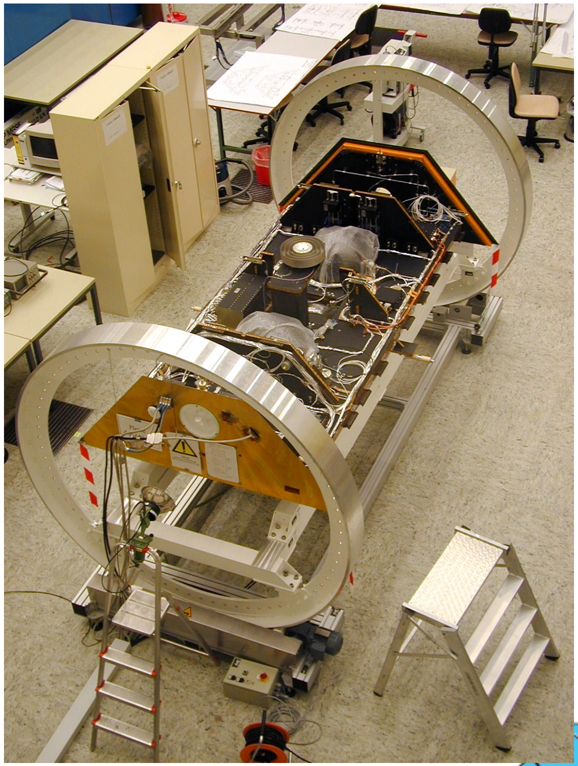
Subsystem Performance Results

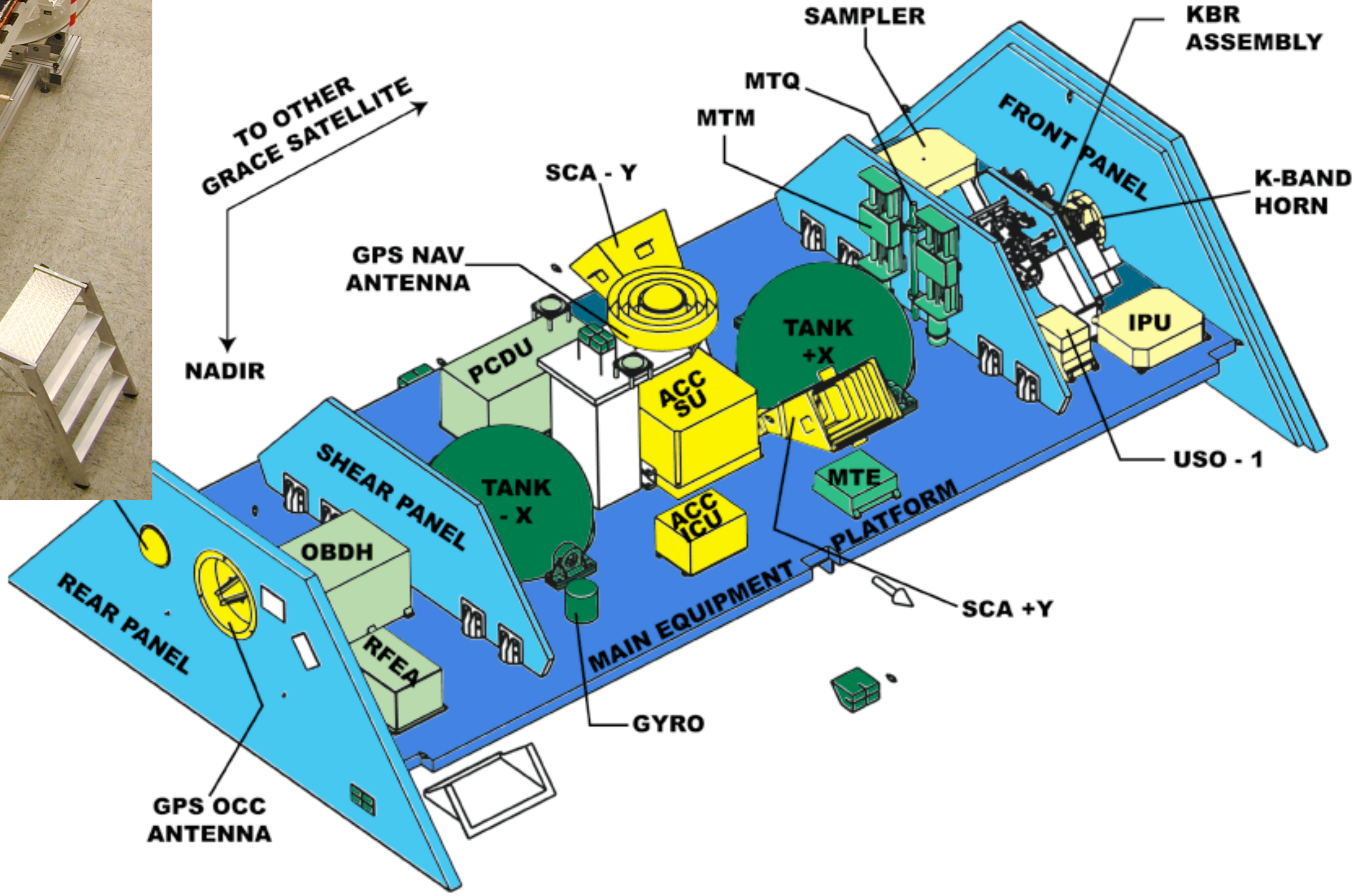


Environmental Test Results

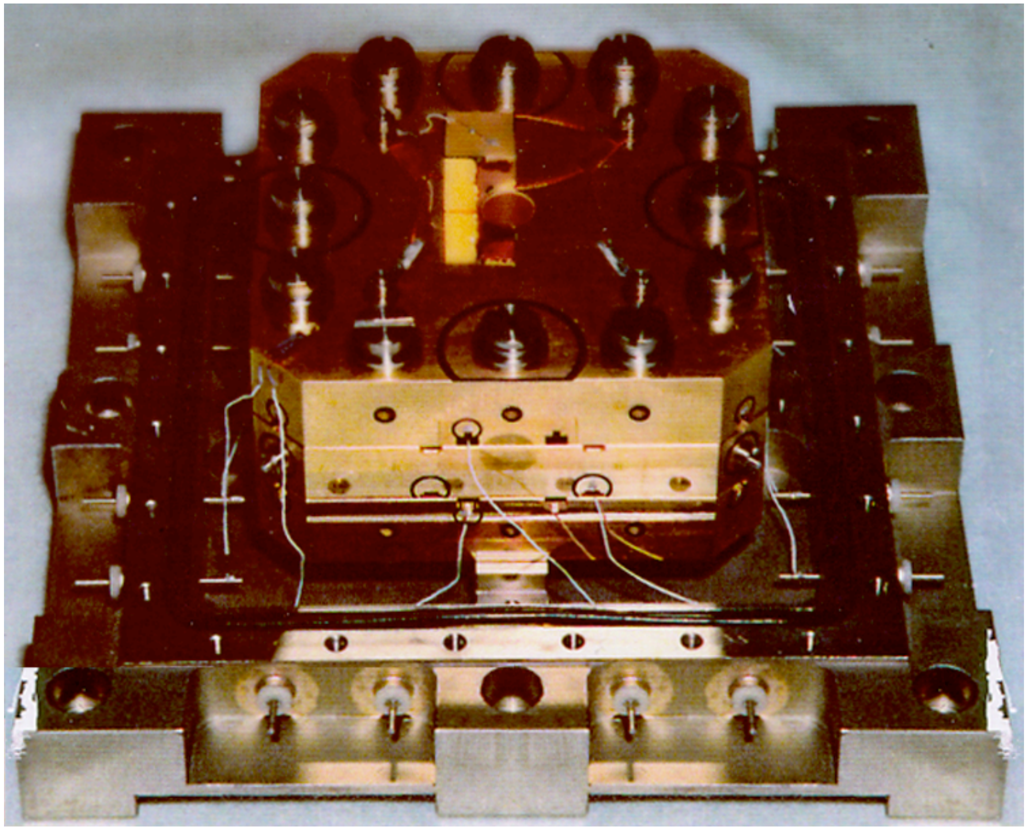
- Breadboard cavity subjected to vibration and thermal tests per the GRACE mission requirements
- Vibration tests were successful
- Thermal tests showed a slight misalignment occurred at highest temperatures (+60 C, +20 C above maximum flight temp.)
  - Traced to adhesive glass transition at 50C
  - Prior uses of same adhesive had not been as sensitive to alignment
  - We are investigating options for prototype unit

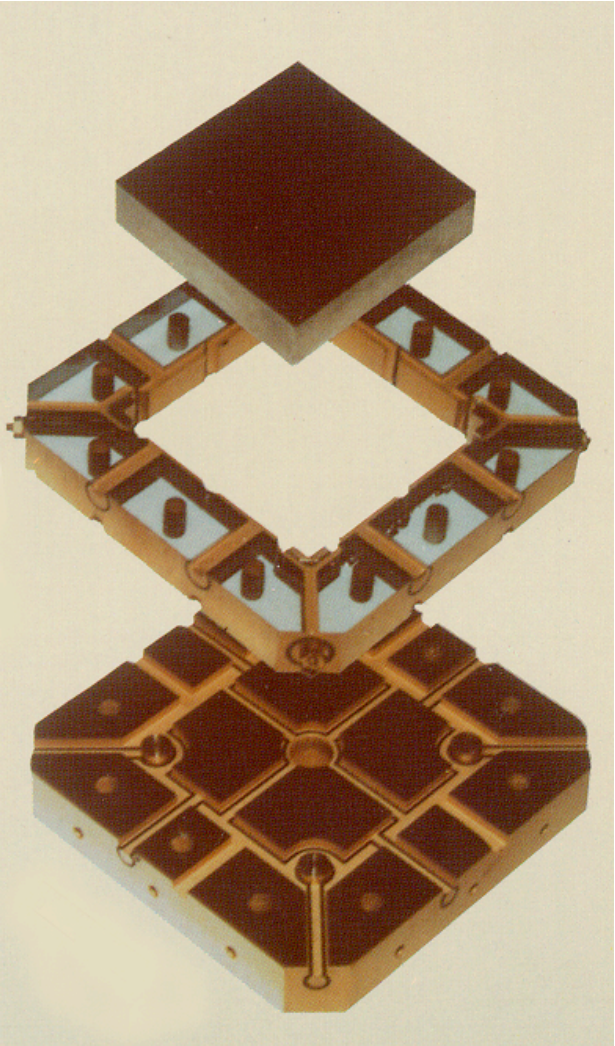
GRACE Spacecraft





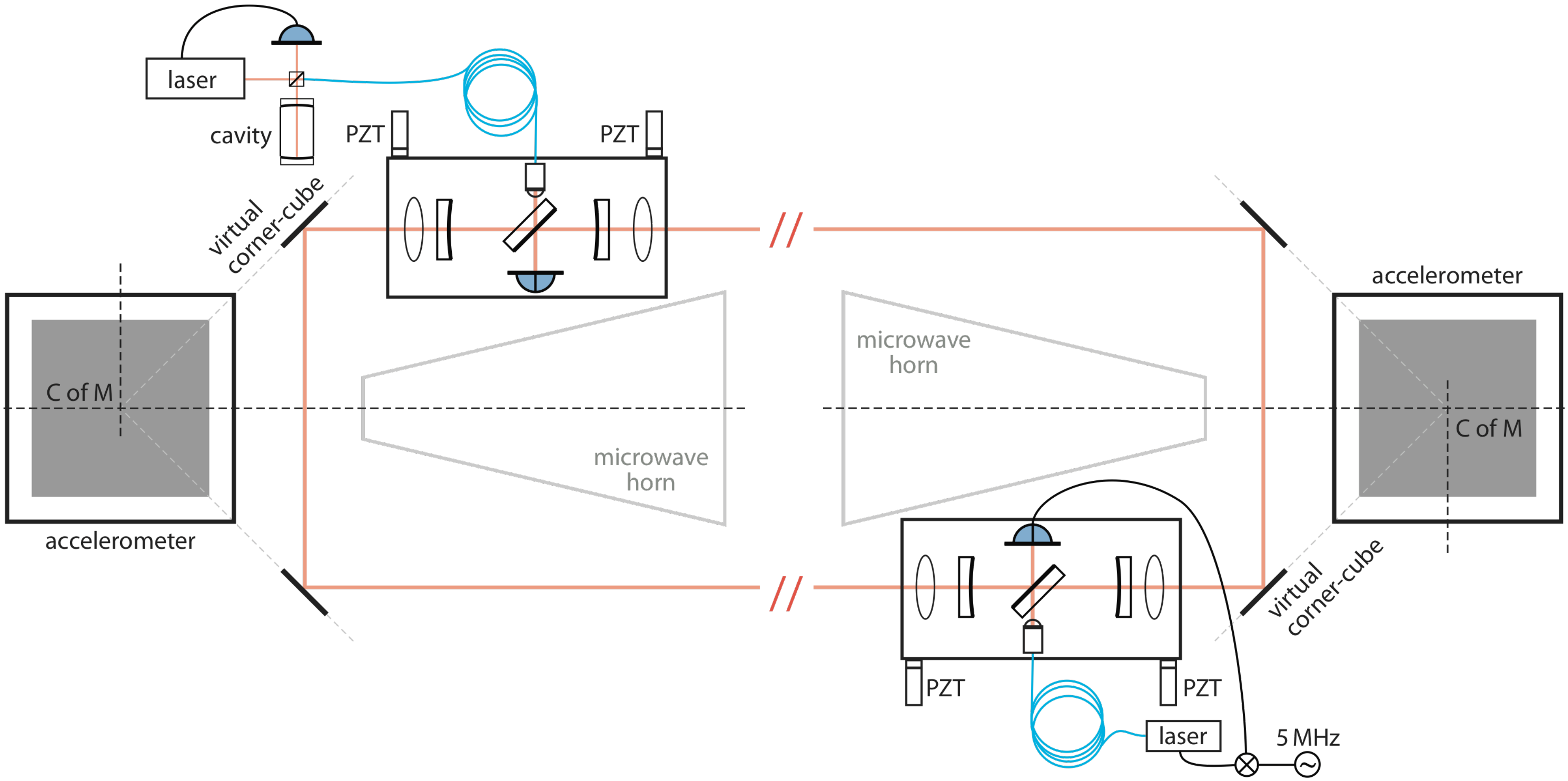
SuperSTAR Accelerometer



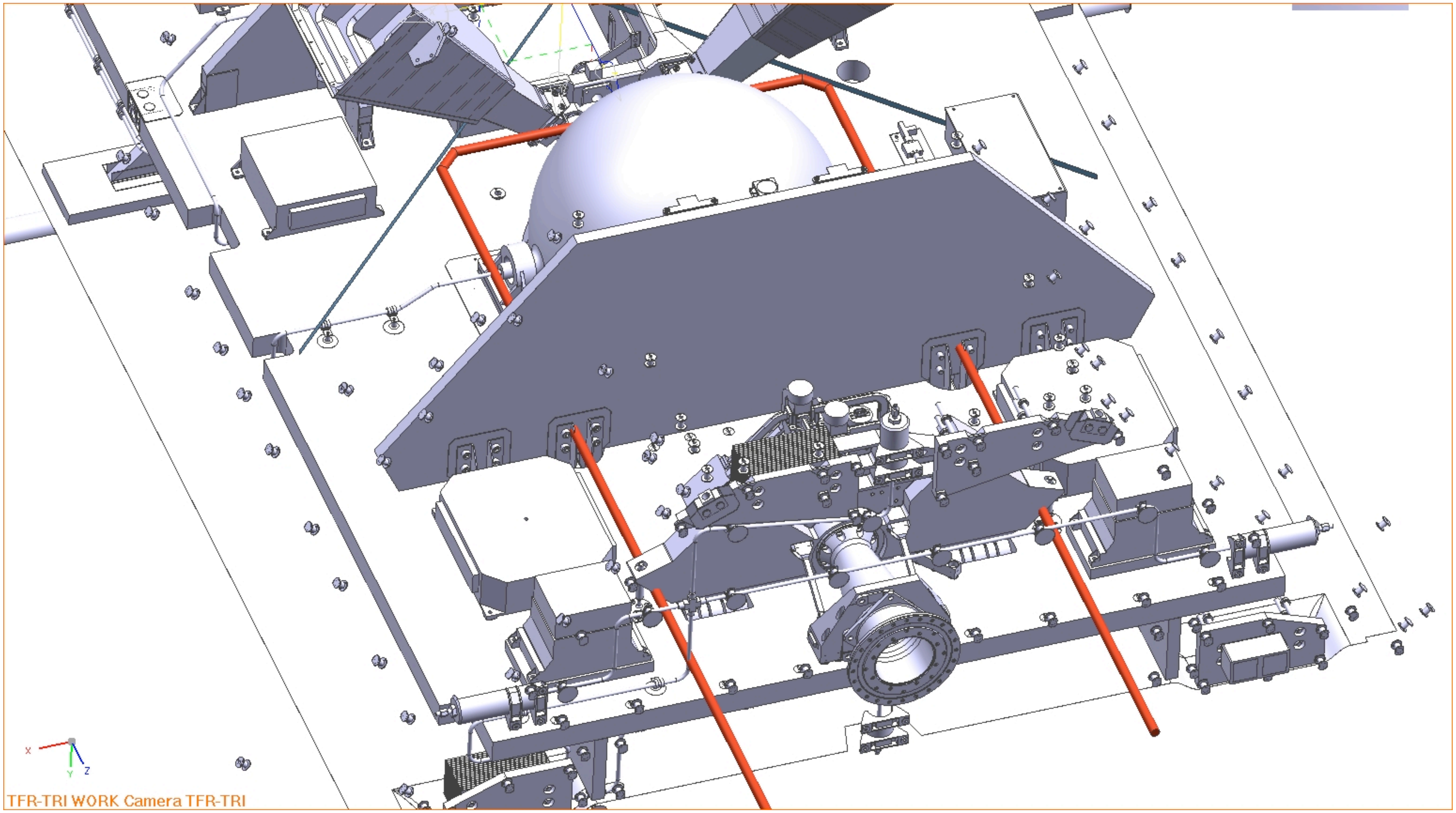


ONERA

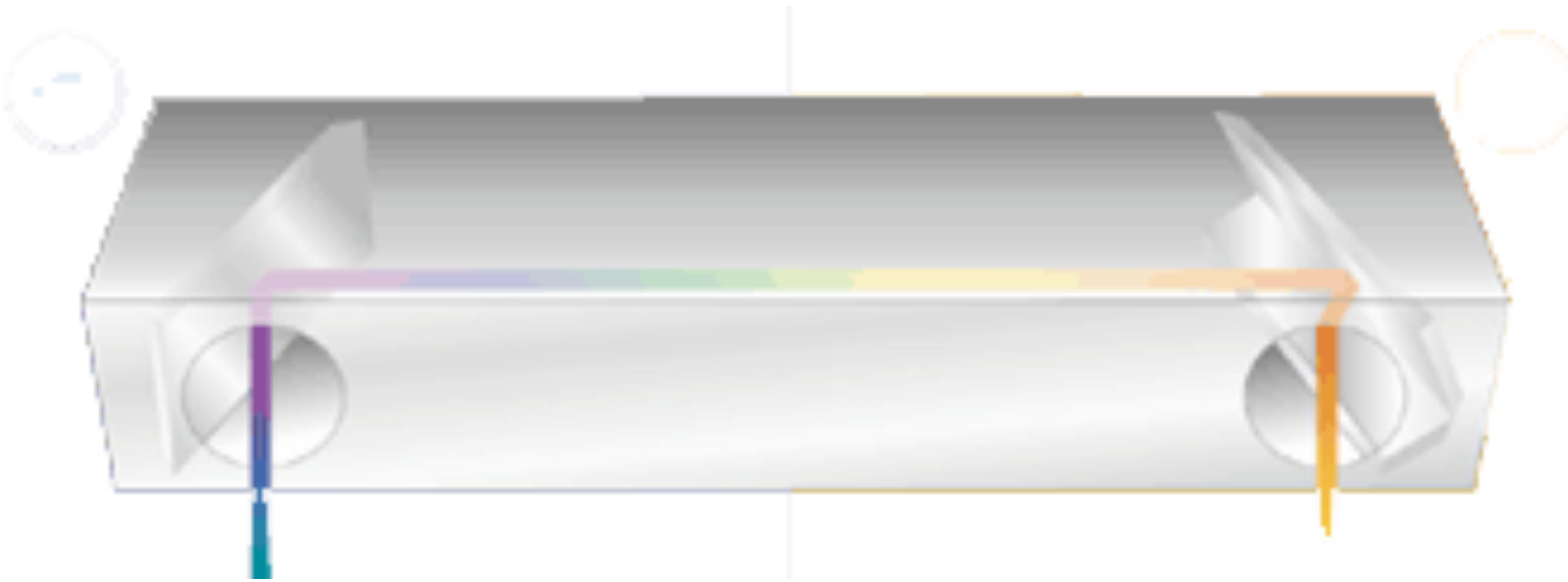
Racetrack Configuration



Laser Optical Path on GRACE-FO

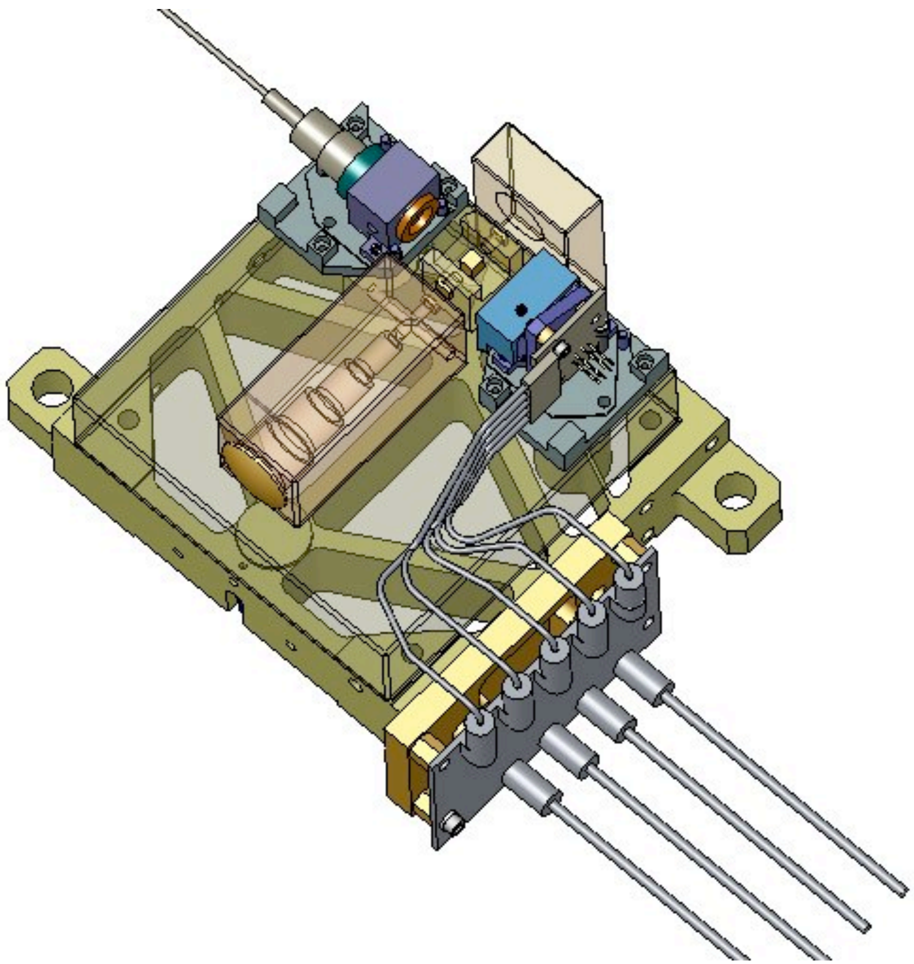


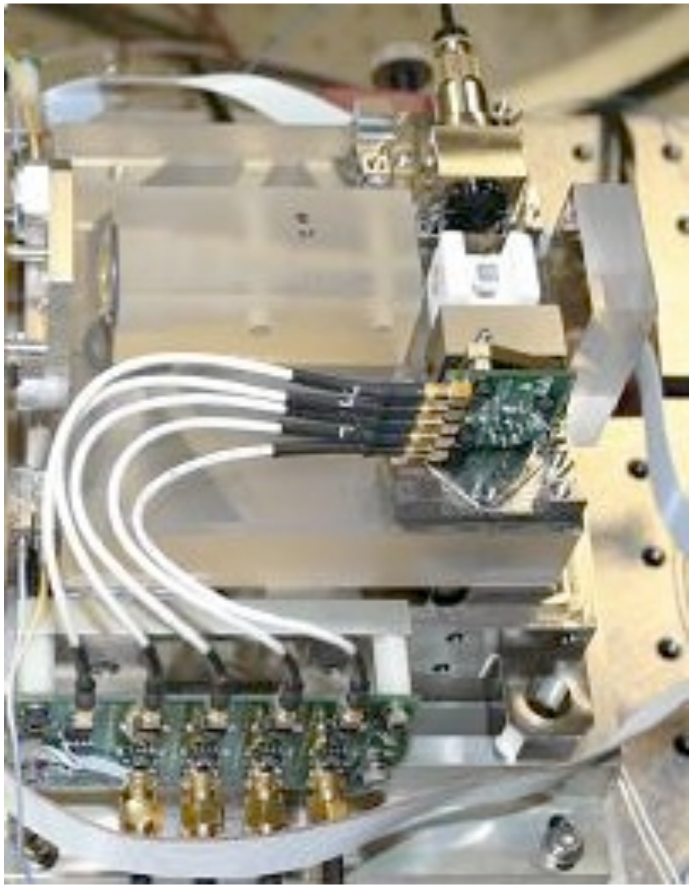
Lateral Transfer Hollow Retroreflector™



PLX

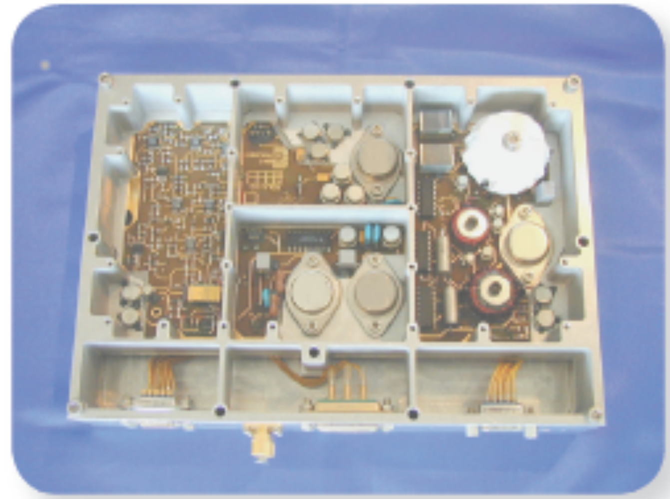
IRT Optical Bench





Candidate PZT Actuators

View of the APAµXS (m = 0.15gr).



FB-LA75-space card.

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Future Plans

- Currently testing performance of modified breadboard subsystem to see if limiting error sources have been identified and corrected
- Prototypes of cavity assembly and electronics board are to be built and tested in FY '11 to demonstrate TRL 6.
- GRACE-FO scheduled to start in FY '11 with PDR in FY'12
  - Cavity and locking electronics development fits GRACE-FO schedule well
  - Will be challenging to bring total ranging system to flight readiness for launch in 2016.